



Vehicle Concept Characteristics - LV 41.4003.08001

UPPER STAGE

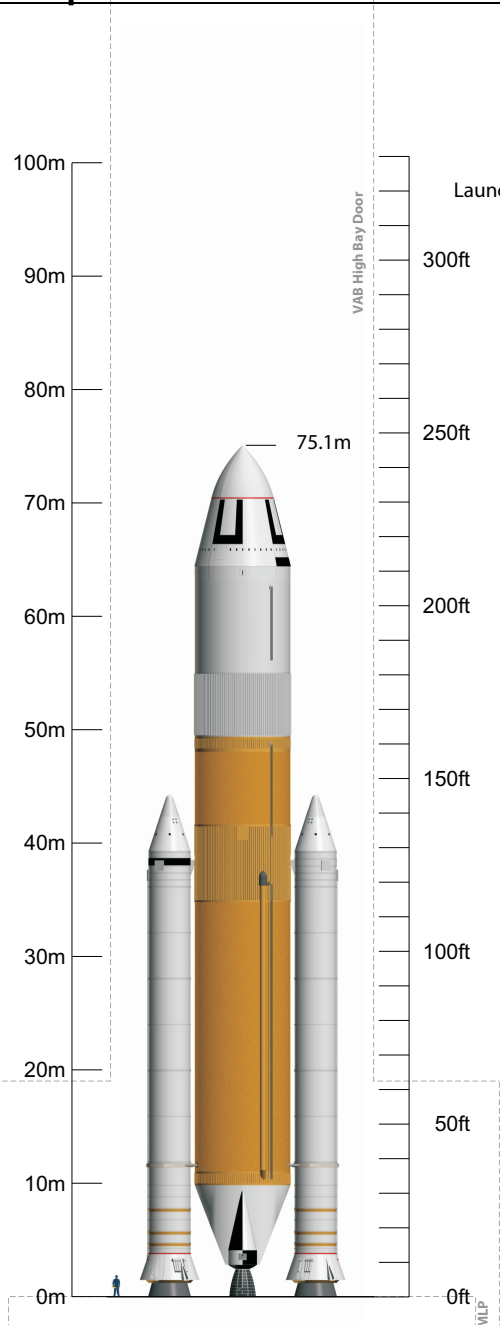
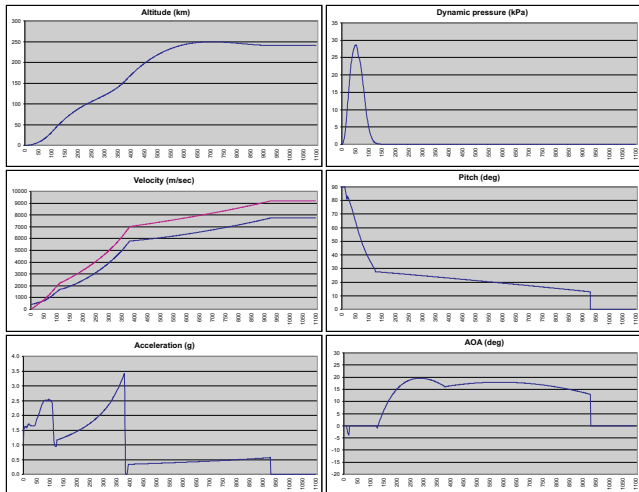
Design Heritage	Boeing ACES / Lockheed-Martin WBC
Propellants	LOX / LH2
Maximum Gross Propellant	385,805lb (174,998kg)
Usable Ascent Propellant	152,399lb (69,127kg)
Ascent Flight Performance Reserve	6,494lb (2,946kg)
Usable Post-Ascent Propellant	220,634lb (100,078kg)
Post-Ascent Flight Performance Reserve	2,229lb (1,011kg)
Unusable Residuals	3,788lb (1,718kg)
Ascent In-Flight Losses	261lb (118kg)
RCS Propellant	992lb (450kg)
Propellant Offload	0.00%
Stage pmf	0.9336
Dry Mass	23,372lb (10,601kg)
Burnout Mass	27,160lb (12,320kg)
# Engines / Type	7 / RL-10A-4-2
Engine Thrust (@ 100%) Vac	22,300lbf (10,115kgf / 99,195N)
Engine Isp (@ 100%) Vac	445.2s
Mission Power Level	100.0%
Upper Stage Ascent Burn Time	541.9s
LEO Loiter Period	4 + 1 days
Pre-TLI Overboard Mass	6,494lb (2,946kg)
ASE*	1,102lb (500kg)

DYNAMICS

Thrust : Weight @ Liftoff	1.534 : 1
Max Dynamic Pressure	599.0psf (28,678Pa)
Max g's During Ascent	3.42g
Insertion Altitude	130.0nmi (240.8km)

ASCENT PERFORMANCE

Delivery Orbit	130.0 x 130.0nmi, 29.0°
Payload w/ regular NASA GR&A's	224,957lb (102,039kg) †
Payload w/ additional 10% Reserve	202,462lb (91,835kg) †



Launch Site

KSC LC-39 (Latitude: 28.6084°)

GLOW

GLOW	4,796,412lb (2,175,616kg)
Payload Fairing	27.6 x 0.0ft (8.4 x 0.0m)
Payload Envelope	25.0 x 0.0ft (7.6 x 0.0m)
Payload Fairing Jettison Mass	8,724lb (3,957kg)
Payload Fairing Jettison	335.9s @ 72.2nmi
Launch Abort System Jettison Mass	-
Launch Abort System Jettison	-

BOOSTERS (each)

Design Heritage	Shuttle RSRM - Flown Unchanged
Propellants	PBAN
Usable Propellant	1,111,604lb (504,215kg)
Stage pmf	0.8561
Dry Mass	183,948lb (83,437kg)
Burnout Mass	186,864lb (84,760kg)
# Boosters / Type	2 / 4-segment Shuttle RSRM
Booster Thrust (@ 0.7s) SL	2,892,912lbf (1,312,203kgf / 12,868,314N)
Vac	3,142,302lbf (1,425,324kgf / 13,977,656N)
Booster Isp (@ 0.7s) SL	237.0s
Vac	269.1s
Booster Burn Time	123.8s

CORE STAGE

Design Heritage	Shuttle Super Light Weight Tank ET
Propellants	LOX / LH2
Gross Propellant	1,621,191lb (735,360kg)
Usable Ascent Propellant	1,604,979lb (728,006kg)
Unusable Residuals	16,047lb (7,279kg)
In-Flight Losses	325lb (147kg)
Propellant Offload	0.00%
Stage pmf	0.9075
Dry Mass	147,479lb (66,895kg)
Burnout Mass	163,526lb (74,174kg)
# Engines / Type	4 / SSME-Block-II
Engine Thrust (@ 104.5%) SL	392,326lbf (177,956kgf / 1,745,155N)
Vac	490,847lbf (222,644kgf / 2,183,396N)
Engine Isp (@ 104.5%) SL	361.4s
Vac	452.2s
Mission Power Level	104.5%
Core Burn Time	384.1s

INTERSTAGE

Dry Mass	8,748lb (3,968kg)
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EDS TLI PERFORMANCE

2-Launch EOR	2-Launch EOR
TLI dV (Adj. for Gravity Losses)	3,205.0m/s (+ FPR)
LEO Loiter Period	5.0 days
TLI Payload Performance*	168,274lb (76,328kg) ‡

Work In Progress

6th June 2009

* ASE is part of the Payload, not additional

† Ascent Performance for Jupiter-247 protects for Upper Stage Single-Engine-Out and full FPR
‡ TLI Performance for Jupiter-247 protects for Upper Stage Dual-Engine-Out and full FPR