



Vehicle Concept Characteristics - LV 41.5002.08001

UPPER STAGE

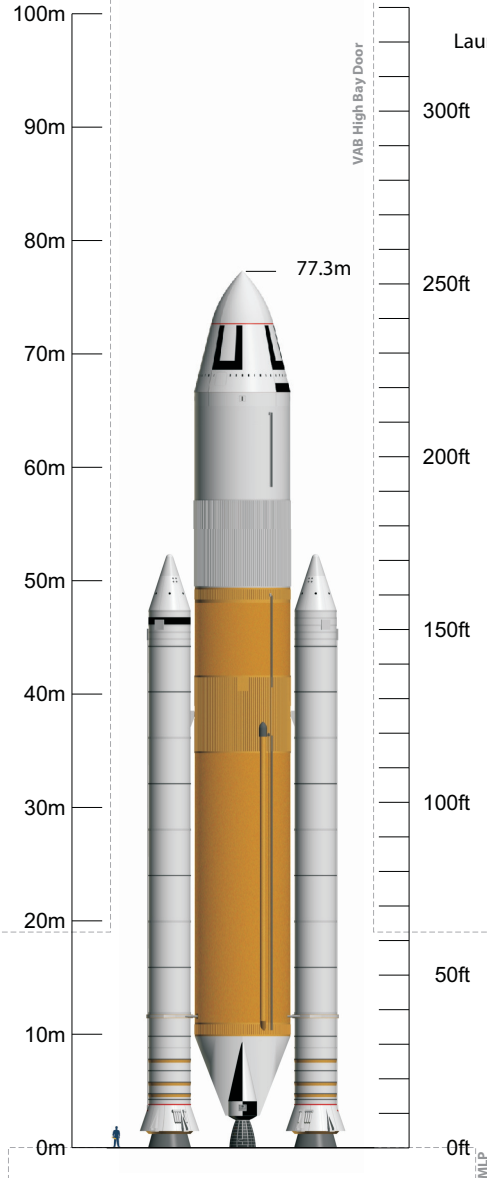
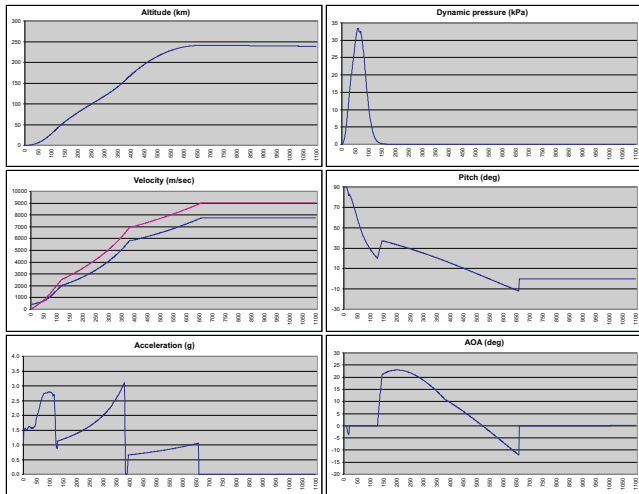
Design Heritage	Boeing ACES / Lockheed-Martin WBC
Propellants	LOX / LH2
Maximum Gross Propellant	438,746lb (199,012kg)
Usable Ascent Propellant	164,736lb (74,723kg)
Ascent Flight Performance Reserve	7,309lb (3,315kg)
Usable Post-Ascent Propellant	259,814lb (117,850kg)
Post-Ascent Flight Performance Reserve	2,624lb (1,190kg)
Unusable Residuals	4,220lb (1,914kg)
Ascent In-Flight Losses	42lb (19kg)
RCS Propellant	992lb (450kg)
Propellant Offload	0.00%
Stage pmf	0.9303
Dry Mass	28,314lb (12,843kg)
Burnout Mass	32,534lb (14,757kg)
# Engines / Type	1 / J-2X
Engine Thrust (@ 100%) Vac	294,000lbf (133,356kgf / 1,307,777N)
Engine Isp (@ 100%) Vac	448.0s
Mission Power Level	100.0%
Upper Stage Ascent Burn Time	271.9s
LEO Loiter Period	4 + 1 days
Pre-TLI Overboard Mass	7,309lb (3,315kg)
ASE*	1,102lb (500kg)

DYNAMICS

Thrust : Weight @ Liftoff	1.512 : 1
Max Dynamic Pressure	698.6psf (33,449Pa)
Max g's During Ascent	3.10g
Insertion Altitude	130.0nmi (240.8km)

ASCENT PERFORMANCE

Delivery Orbit	130.0 x 130.0nmi, 29.0°
Payload w/ regular NASA GR&A's	264,533lb (119,990kg)
Payload w/ additional 10% Reserve	238,079lb (107,991kg)



Launch Site

KSC LC-39 (Latitude: 28.6084°)

GLOW

GLOW	5,750,543lb (2,608,402kg)
Payload Fairing	27.6 x 0.0ft (8.4 x 0.0m)
Payload Envelope	25.0 x 0.0ft (7.6 x 0.0m)
Payload Fairing Jettison Mass	8,724lb (3,957kg)
Payload Fairing Jettison	335.5s @ 72.5nmi
Launch Abort System Jettison Mass	-
Launch Abort System Jettison	-

BOOSTERS (each)

Design Heritage	Shuttle-derived 5-segment RSRMV
Propellants	PBAN
Usable Propellant	1,380,873lb (626,353kg)
Stage pmf	0.8656
Dry Mass	228,620lb (103,700kg)
Burnout Mass	232,608lb (105,509kg)
# Boosters / Type	2 / 4-segment Shuttle RSRM
Booster Thrust (@ 0.7s) SL	3,510,791lbf (1,592,468kgf / 15,616,776N)
Vac	3,510,791lbf (1,592,468kgf / 15,616,776N)
Booster Isp (@ 0.7s) SL	237.0s
Vac	267.4s
Booster Burn Time	126.6s

CORE STAGE

Design Heritage	Shuttle Super Light Weight Tank ET
Propellants	LOX / LH2
Gross Propellant	1,621,191lb (735,360kg)
Usable Ascent Propellant	1,604,979lb (728,006kg)
Unusable Residuals	16,047lb (7,279kg)
In-Flight Losses	325lb (147kg)
Propellant Offload	0.00%
Stage pmf	0.9075
Dry Mass	147,479lb (66,895kg)
Burnout Mass	163,526lb (74,174kg)
# Engines / Type	4 / SSME-Block-II
Engine Thrust (@ 104.5%) SL	392,326lbf (177,956kgf / 1,745,155N)
Vac	490,847lbf (222,644kgf / 2,183,396N)
Engine Isp (@ 104.5%) SL	361.4s
Vac	452.2s
Mission Power Level	104.5%
Core Burn Time	384.1s

INTERSTAGE

Dry Mass	11,664lb (5,291kg)
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EDS TLI PERFORMANCE

2-Launch EOR	
TLI dV (Adj. for Gravity Losses)	3,175.0m/s (+ FPR)
LEO Loiter Period	5.0 days
TLI Payload Performance*	203,066lb (92,109kg)

* ASE is part of the Payload, not additional